

Square Root of $\begin{bmatrix} \mathbf{T}_q & \mathbf{0} \\ \mathbf{0} & 1 \end{bmatrix}$

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Let \mathbf{q} be a unit quaternion. Its inverse is its conjugate \mathbf{q}^* . Recall the quaternion operators \otimes and \circledast and the $[\mathbf{q}\otimes]$ and $[\mathbf{q}\circledast]$ operator matrices, which were defined in [1, App.] and elsewhere. For quaternions \mathbf{p} and \mathbf{r} , we have $\mathbf{r} = [\mathbf{q}\otimes]\mathbf{p} = \mathbf{q}\otimes\mathbf{p}$ and $\mathbf{r} = [\mathbf{p}\circledast]\mathbf{q} = \mathbf{p}\circledast\mathbf{q}$.

Define the matrix

$$C = \begin{bmatrix} -\mathbf{I} & \mathbf{0} \\ \mathbf{0} & 1 \end{bmatrix} \quad (1)$$

Note that $C^2 = I$. We could call C a conjugation operator matrix since $\mathbf{q}^* = C\mathbf{q}$. It is easily verified that

$$C[\mathbf{q}\otimes]C = [\mathbf{q}^*\circledast] = [\mathbf{q}\circledast]^T \quad (2)$$

from which we have that

$$C[\mathbf{q}\otimes] = [\mathbf{q}\circledast]^T C \quad (3a)$$

and

$$[\mathbf{q}\otimes]C = C[\mathbf{q}\circledast]^T \quad (3b)$$

It is well known that

$$[\mathbf{q}\circledast]^T[\mathbf{q}\otimes] = [\mathbf{q}\otimes][\mathbf{q}\circledast]^T = \begin{bmatrix} \mathbf{T}_q & \mathbf{0} \\ \mathbf{0} & 1 \end{bmatrix} \quad (4)$$

where \mathbf{T}_q is the direction cosine or attitude matrix that corresponds to the quaternion \mathbf{q} .

From the foregoing relations, we conclude that $C[\mathbf{q}\otimes]$ and $[\mathbf{q}\otimes]C$ are two square roots of $\begin{bmatrix} \mathbf{T}_q & \mathbf{0} \\ \mathbf{0} & 1 \end{bmatrix}$ because, from (3a) and (4), we have

$$\begin{aligned} (C[\mathbf{q}\otimes])^2 &= (C[\mathbf{q}\otimes])(C[\mathbf{q}\otimes]) \\ &= C[\mathbf{q}\otimes]C[\mathbf{q}\otimes] \\ &= [\mathbf{q}\circledast]^T[\mathbf{q}\otimes] \\ &= \begin{bmatrix} \mathbf{T}_q & \mathbf{0} \\ \mathbf{0} & 1 \end{bmatrix} \end{aligned} \quad (5a)$$

and similarly from (3b) and (4)

$$\begin{aligned} ([\mathbf{q}\otimes]C)^2 &= ([\mathbf{q}\otimes]C)([\mathbf{q}\otimes]C) \\ &= [\mathbf{q}\otimes]C[\mathbf{q}\otimes]C \\ &= [\mathbf{q}\otimes][\mathbf{q}\circledast]^T \\ &= \begin{bmatrix} \mathbf{T}_q & \mathbf{0} \\ \mathbf{0} & 1 \end{bmatrix} \end{aligned} \quad (5b)$$

Define

$$\mathbf{q} = \begin{bmatrix} \mathbf{e} \\ c \end{bmatrix} \quad [\mathbf{q}\otimes] = \begin{bmatrix} c\mathbf{I} - [\mathbf{e}\times] & \mathbf{e} \\ -\mathbf{e}^T & c \end{bmatrix} \quad (6)$$

where \mathbf{e} is the vector part of the quaternion and c is the scalar part. Then from the (5b), we have

$$\mathbf{T}_q = \begin{bmatrix} -c\mathbf{I} + [\mathbf{e}\times] & \mathbf{e} \\ \mathbf{e}^T & \end{bmatrix} \begin{bmatrix} -c\mathbf{I} + [\mathbf{e}\times] \\ \mathbf{e}^T \end{bmatrix} \quad (7)$$

If we start from (5a), the result differs only in a sign that cancels out. We have, then,

$$\mathbf{T}_q = (c\mathbf{I} - [\mathbf{e}\times])^2 + \mathbf{e}\mathbf{e}^T \quad (8a)$$

$$= \mathbf{I} - 2c[\mathbf{e}\times] + 2[\mathbf{e}\times]^2 \quad (8b)$$

Equation (8b) is a common expression for \mathbf{T}_q in terms of \mathbf{q} , whereas (8a) is a (new?) quadratic form. The quadratic form could be used to transform a vector via the following steps

$$\mathbf{v}' = c\mathbf{v} - \mathbf{e} \times \mathbf{v}$$

$$\mathbf{v}'' = c\mathbf{v}' - \mathbf{e} \times \mathbf{v}' + \mathbf{e}(\mathbf{e}^T \mathbf{v})$$

so that $\mathbf{v}'' = \mathbf{T}_q \mathbf{v}$.

Observe that a factorization of \mathbf{T}_q can be obtained from a QR decomposition, viz

$$\begin{bmatrix} -c\mathbf{I} + [\mathbf{e}\times] \\ \mathbf{e}^T \end{bmatrix} = QR = [Q_1 \ Q_2] \begin{bmatrix} R_1 \\ \mathbf{0} \end{bmatrix} = Q_1 R_1 \quad (9)$$

where Q is orthogonal and R_1 is upper triangular. A factorization of \mathbf{T}_q is then

$$\mathbf{T}_q = R_1^T R_1 \quad (10)$$

Note that the process of computing R_1 is essentially one of completing the square in equation (8a).

Acknowledgement

Landis Markley pointed out that the matrix square root was derived by Bernard Friedland [2].

References

- [1] Pittelkau, M. E., "Measurement Sensitivity Equations in Attitude Determination", *2003 Flight Mechanics Symposium*, NASA Goddard Space Flight Center, NASA CP-2003-212246, 28–30 Oct. 2003.
- [2] Bernard Friedland, "Analysis Strapdown Navigation Using Quaternions," *IEEE AES-14*, No. 5, 1978, pp. 764–768.